

SIRIUS: Stellar and ISM research via in-orbit (E)UV spectroscopy

M.A. Barstow

(on behalf of a consortium from the EU, NAOC,
Peking and Hefei)

Introduction

- Mission opportunity gap
 - From ~€1M to ~€200M
- A scale-able high throughput spectrograph
- Observatory-class science goals from a small mission
- Readiness levels and collaboration opps.
- Conclusions



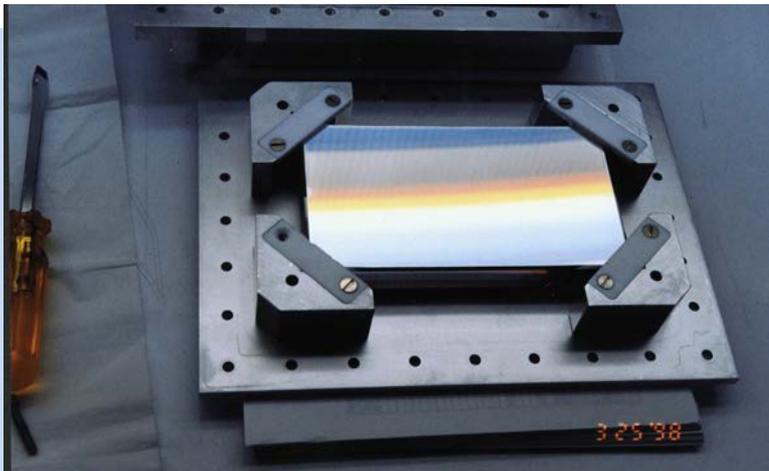
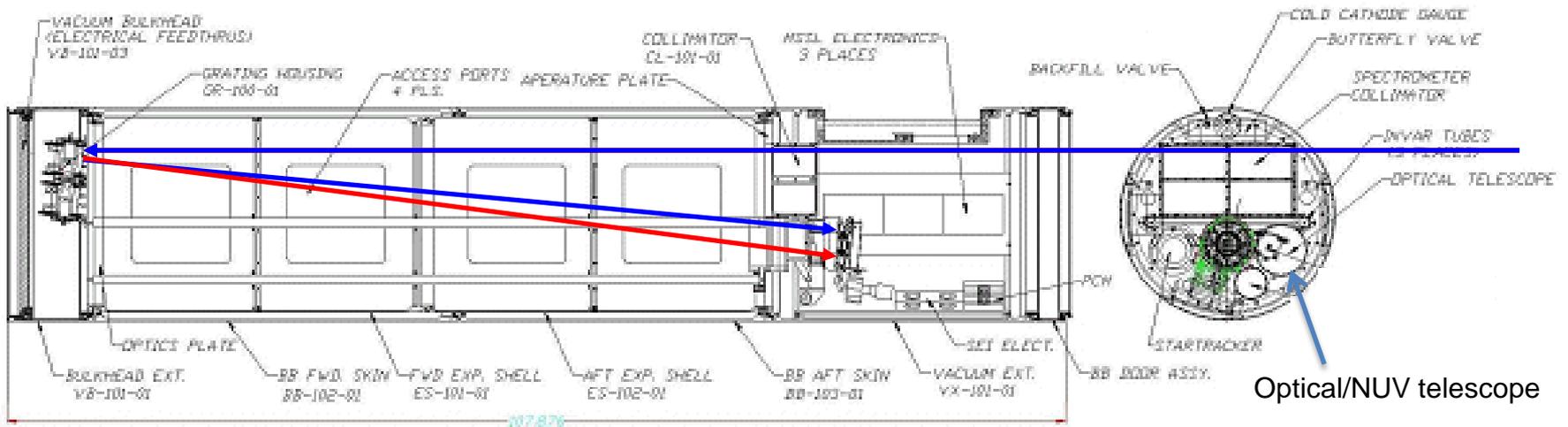
Mission opportunity gap

- Sounding rockets (WFSXC, J-PEX)
 - ~5 min... limited S/N, timing etc.
 - ~€2-3M (~€5k/sec; €2M/target)
 - Frequent flights
- Orbital (EXOSAT, ROSAT, EUVE, Hubble, FUSE)
 - 3 to 24 years
 - ~€200-2000M (~€2/sec; €50k/target)
 - Few opportunities
- Need something in between

Filling the gap - small missions

- Focused science goals - something new
- Control over requirement inflation
- Imaginative use of new technology
- But high TRL concepts
- Modular payloads, standard components & subsystems
- Innovative ground-station & mission operations concepts

J-PEX Normal Incidence Spectrometer



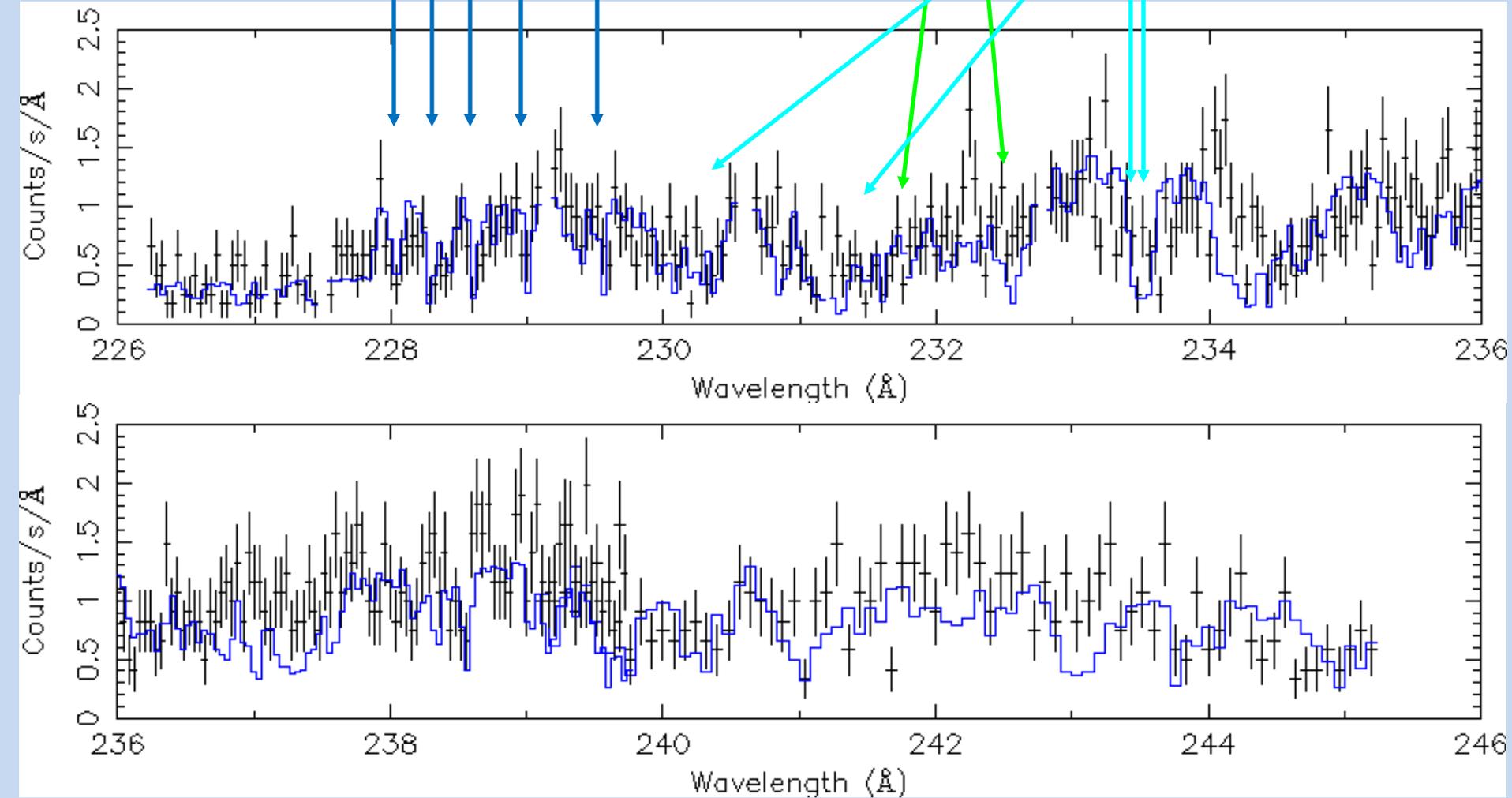
Ion etched, blazed grating.
MoSi multilayers for high reflection
Spherical figure, 2.2m focal length



Helium II

Nitrogen IV

Oxygen IV

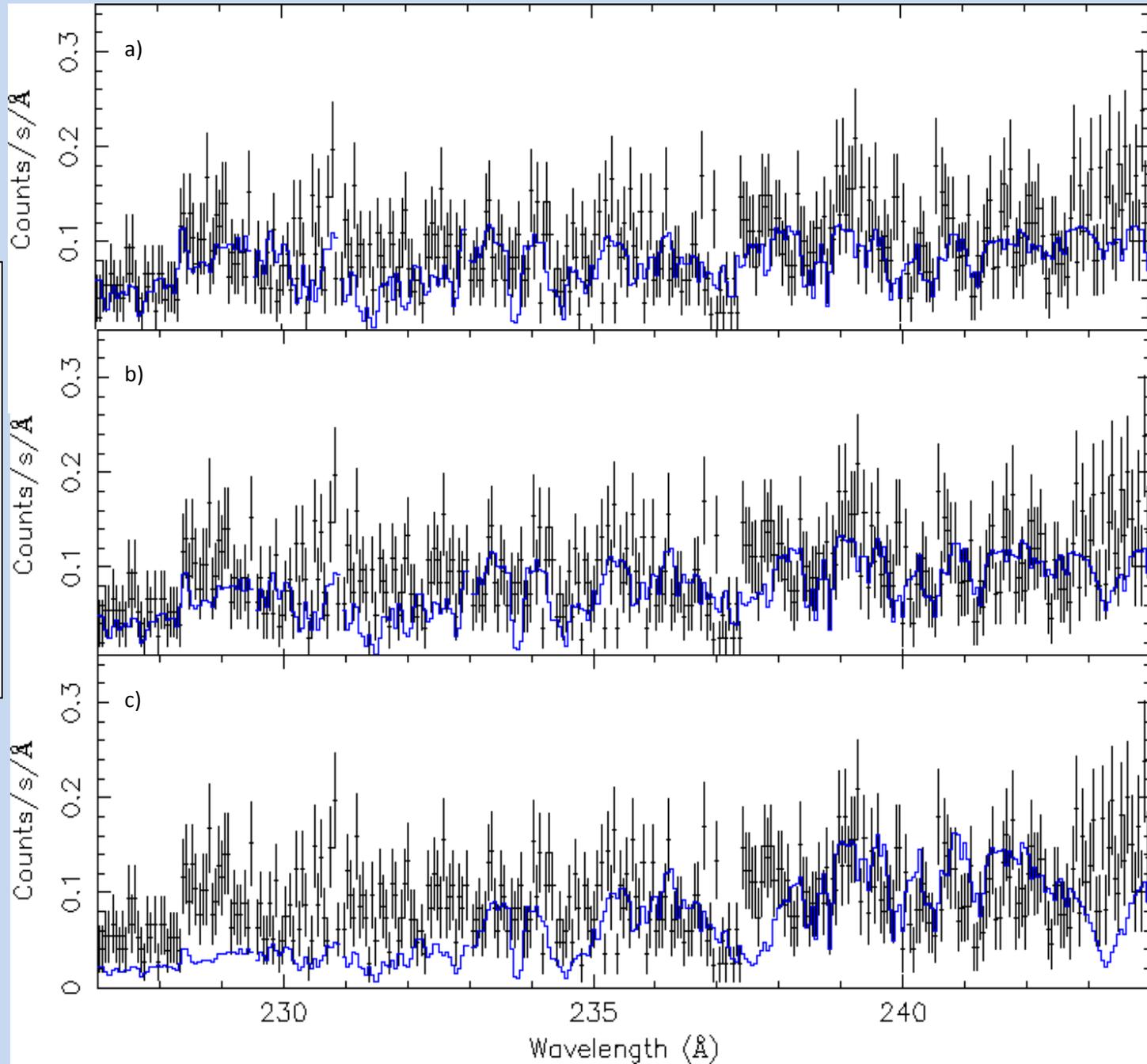


Feige 24; DA+dM

a) H-layer mass =
 $10^{-12.92} M_{\odot}$

b) H-layer mass =
 $10^{-13.5} M_{\odot}$

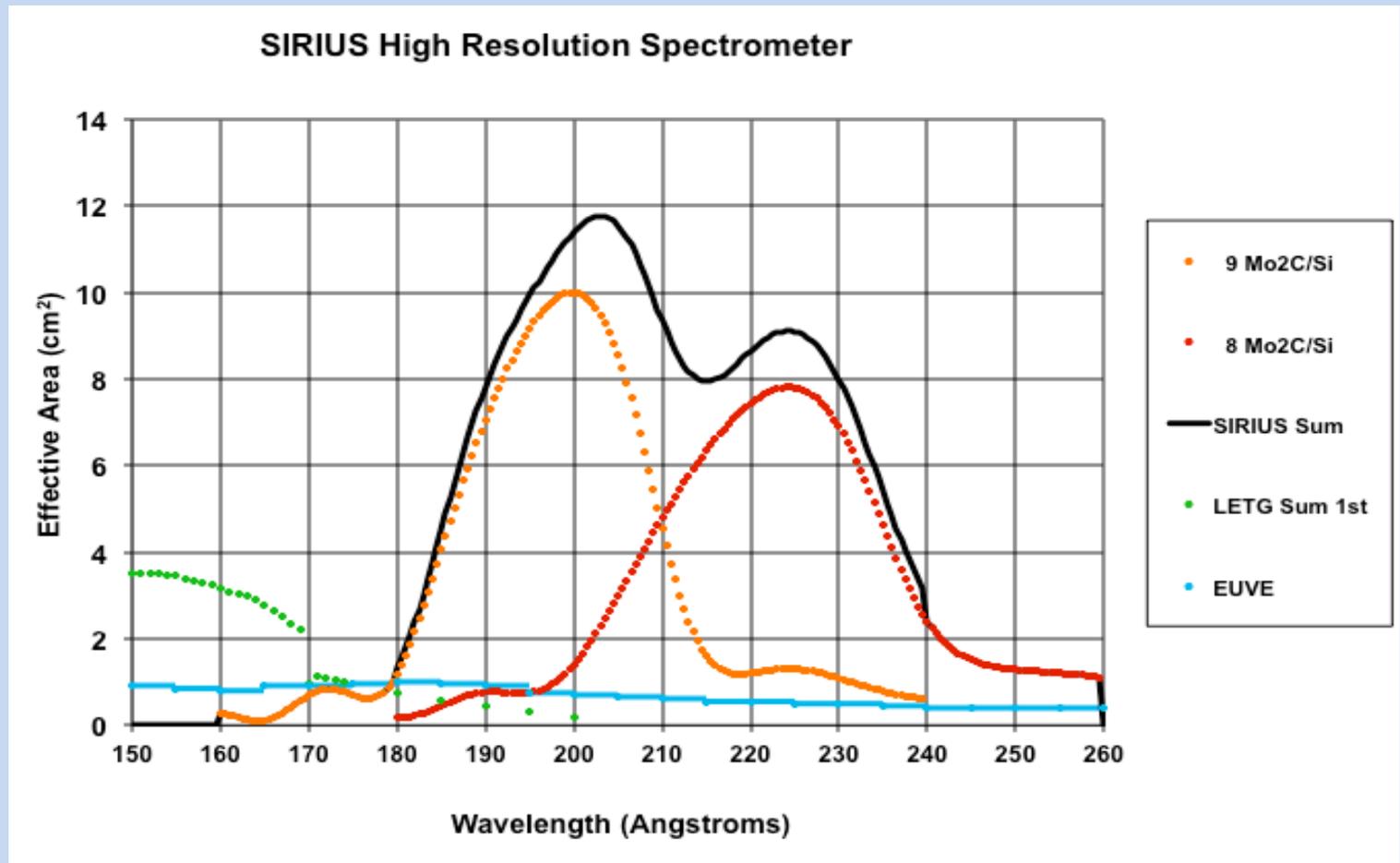
c) H-layer mass =
 $10^{-14} M_{\odot}$



Evolution into an orbital instrument

- The J-PEX design is scale-able
- Multiple units can be constructed
 - Tuned to different wavebands by multilayer thickness
- Gratings in a single unit can be divided for broader wavelength coverage
 - Recent, shortlisted, ESA S-mission proposal

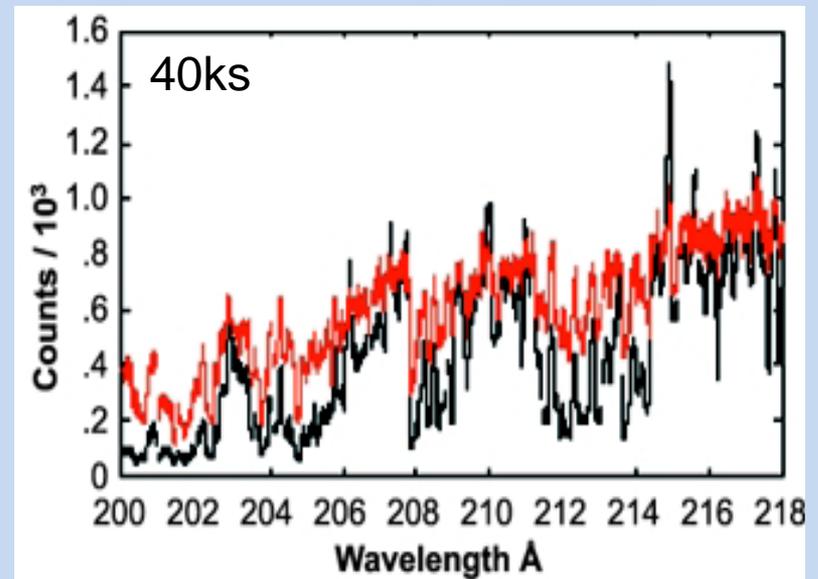
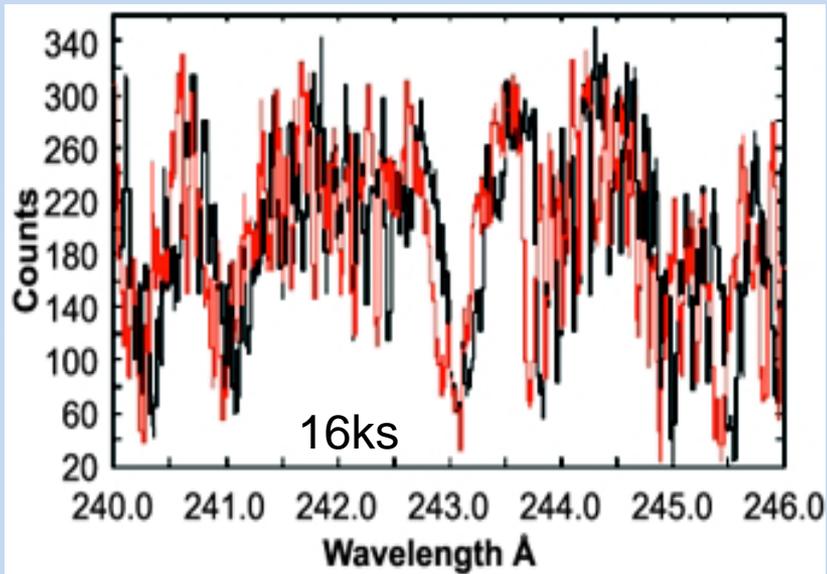
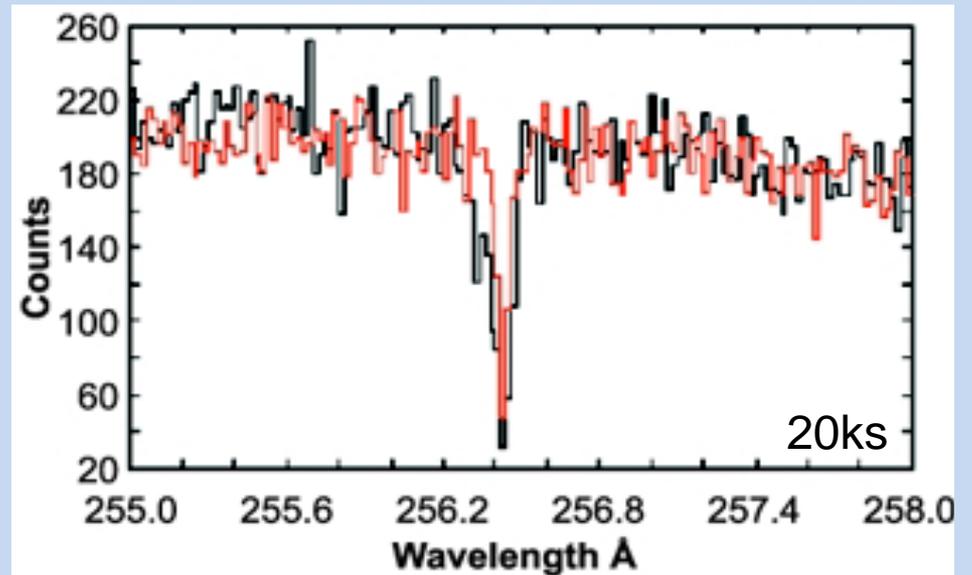
SIRIUS

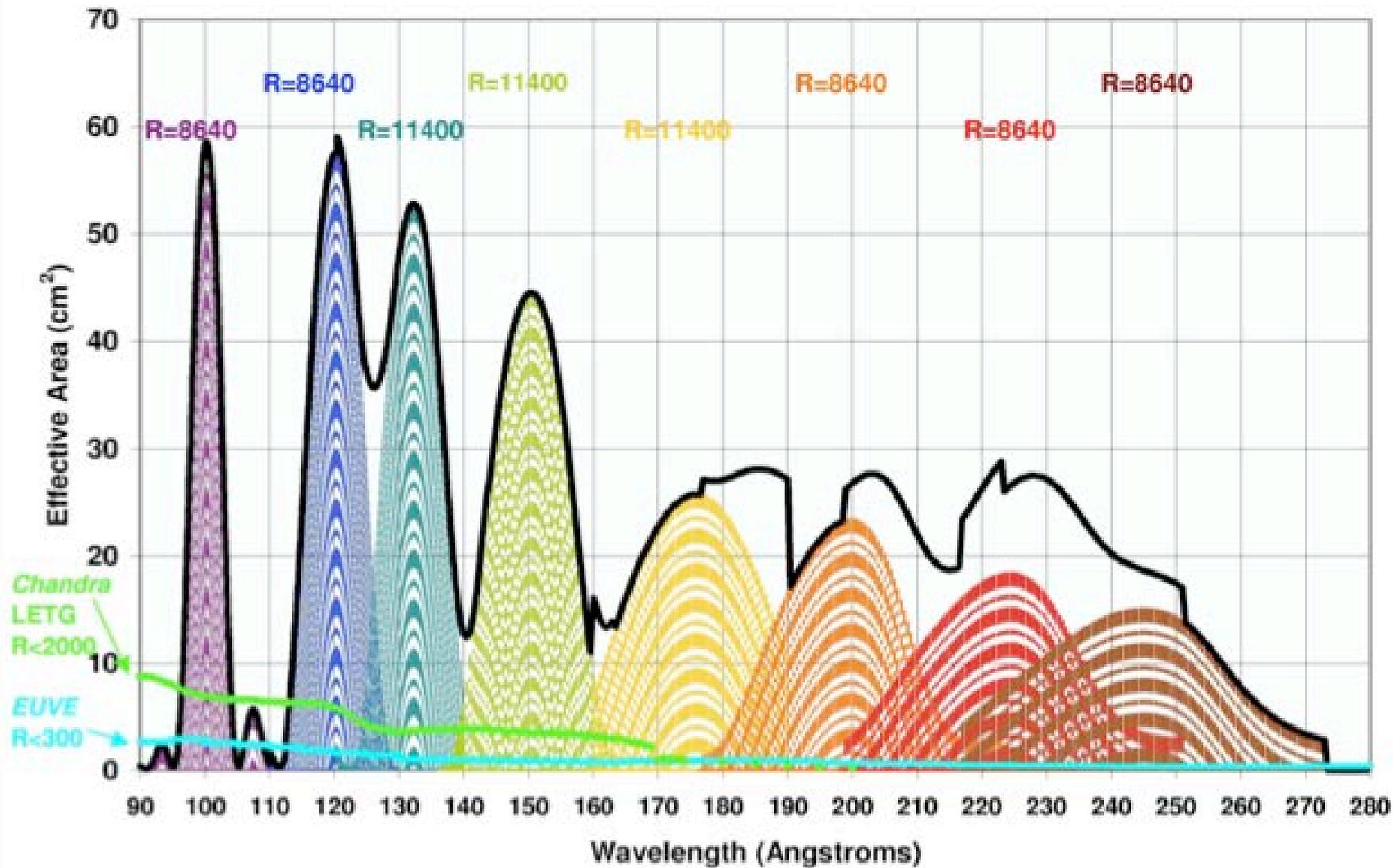


Science - The Galactic Environment

- Structure & Dynamics of Stellar Coronae
 - Coronal Activity - Solar quality data for nearby stars
 - Exoplanet Environments
- Evolution of White Dwarfs
 - Atmospheric composition & structure
 - Extrasolar planetary debris (arXiv - 1402.2164)
- Structure & Ionization of the Local Interstellar Gas
 - Can only be directly measured in the EUV

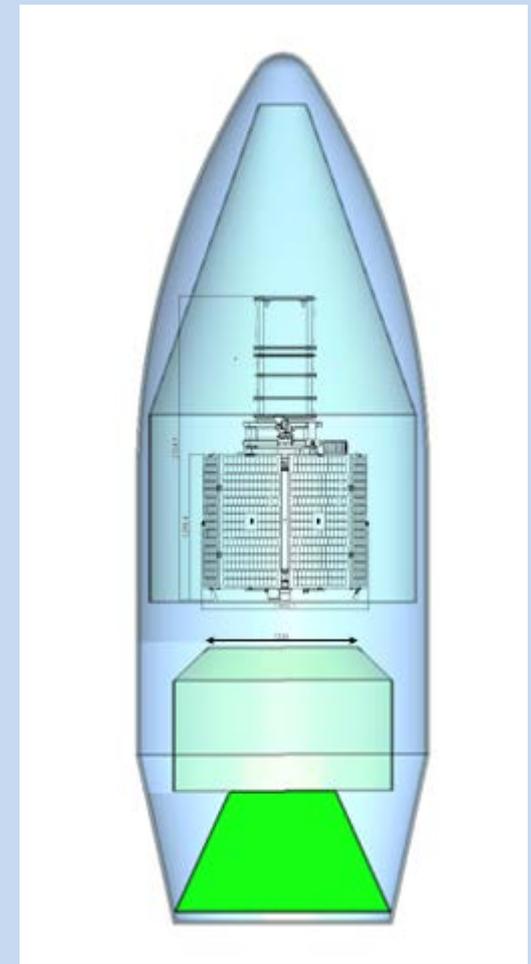
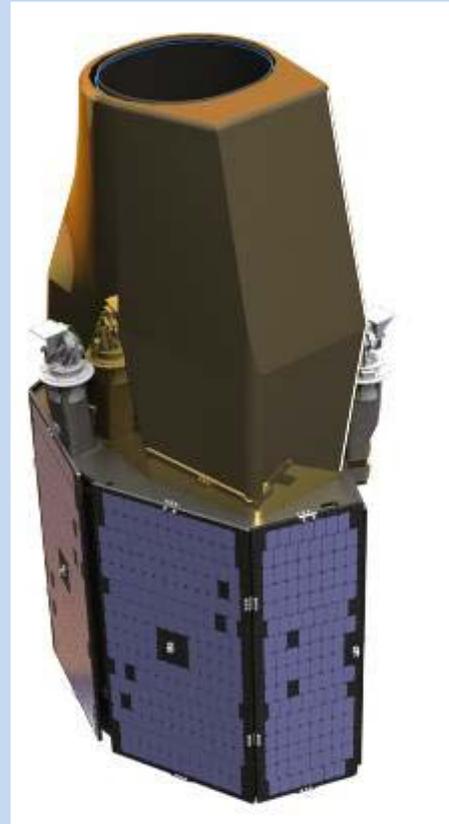
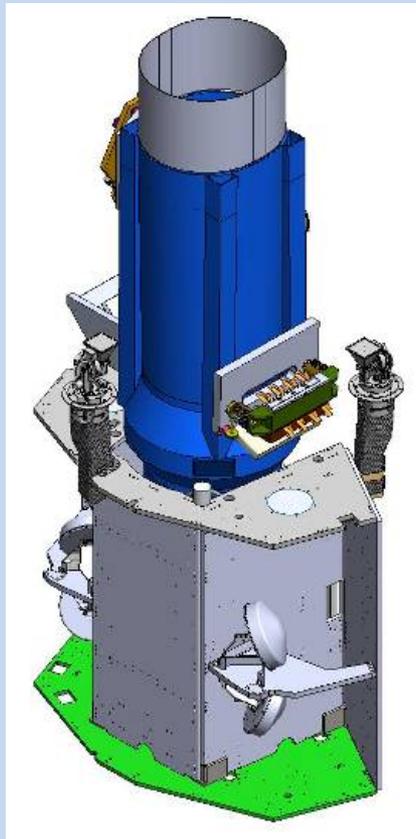
Capability



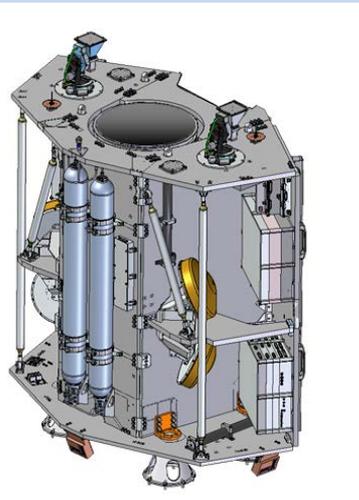


Assembled Satellite

SSTL 300



Vega Launch



Mission Summary

Category	Requirements
Launch Vehicle	Shared launch on Vega or Long March 2C/2D
Trajectory Design	Science Orbit: LEO, above 300km
Flight System	Design Lifetime: 5 years, nominal mission 3 years Consumables: 5 years Payload: EUV spectrograph + opt/UV monitor Mass:55 kg, Power:13.5 W Observing Mode: 3-axis Platform Pointing Accuracy: 1 arcmin Pointing Stability: 1 arcsec/s Average Science Data Acquisition Rate: 20kb/s
Spacecraft Bus	SSTL-300 or Chinese
Data Return Strategy	Downlink Volume <1.2Gb per day, X-band
Telecommunications	ESA, China or UKSA
Ground Systems	2 contacts per day
Solar Array	1.66 m ² Emcore BTJM triple junction GaAs cells
Propulsion (for attitude control)	24kg Xe, 32m/s delta-v at an Isp of 48s

Technology opportunities & readiness

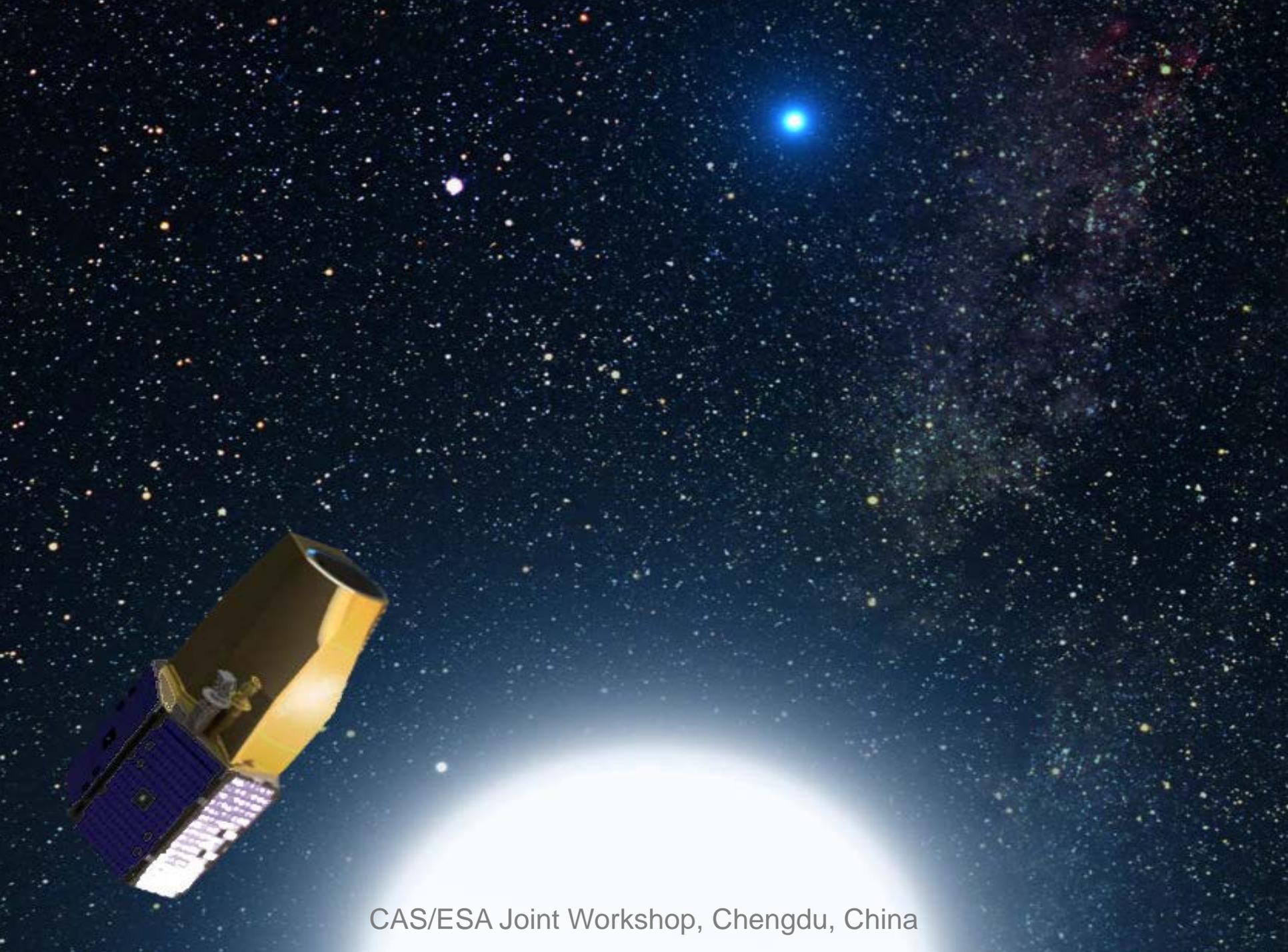
- Opportunities for the EU
 - Gratings (TRL \geq 7) - Carl Zeiss, Horiba
 - Detectors (TRL \geq 7) - EU institutes, e2v, Photonis
 - Spacecraft - SSTL, Astrium
 - Launch, ground segment & operations
- Opportunities for China
 - Optical/NUV telescope
 - Structure, collimator, processing electronics
 - Spacecraft - TBD
 - Launch, ground segment & operations

CAS/ESA Joint Workshop, Chengdu, China



Conclusion

- New technology & innovative design yield high sensitivity instrumentation
- New, unique science from a unique instrument
- Observatory-class science at low cost
- Study indicates our proposed SIRIUS EUV spectrograph feasible (within the defined parameters) for this joint opportunity
- Happy to discuss potential ways forward with ESA/CAS colleagues



CAS/ESA Joint Workshop, Chengdu, China